



AEROSPACE STANDARD	AS269™	REV. J
	Issued 1954-12 Revised 1996-09 Reaffirmed 2012-11 Stabilized 2018-08	
Superseding AS269H		
Identification of Material for AN, MS, and AS Engine and Propeller Standard Utility Parts and Also for Company Parts		

RATIONALE

This document has been determined to contain basic and stable technology which is not dynamic in nature.

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1. SCOPE:

This SAE Aerospace Standard (AS) provides material identification codes for aluminum alloys, copper alloys, carbon steels, alloy steels, titanium alloys, corrosion resistant materials, and heat resistant materials that are used to make AN, MS, and AS engine and propeller standard utility parts. This document also provides similar material codes for company parts (such as nuts, bolts, etc.) having design configuration similar to other company parts or the engine and propeller standard utility parts (AN, MS, and AS) that are not of the same material.

2. REFERENCES:

All documents listed in Figure 1 and Table 1 are available from SAE, 400 Commonwealth Drive, Warrendale, PA 15096-0001.

3. MATERIAL IDENTIFICATION CODE:

3.1 Individual material code designations are restricted to materials of the same composition, irrespective of the raw stock configuration.

3.2 AN, MS, and AS Engine and Propeller Standard Utility Parts:

The material codes established for engine and propeller standard utility parts (AN, MS, and AS) are prefixed with the letter "E" in order to identify engine and propeller constant quality standard parts. In common temperature ranges the "E" prefix and number comprise the material code for materials specified on parts generally used in temperature applications not exceeding 550 °F (288 °C). Corrosion resistant materials for parts generally used in temperature applications between 550 °F and 800 °F (288 °C and 427 °C) have a material code consisting of an "EC" prefix and a number. Heat resistant materials for parts generally used in temperature applications above 800 °F (427 °C) have a material code consisting of an "EH" prefix and a number. Titanium alloys have a material code consisting of an "ET" prefix and a number.

3.3 Company Parts:

The material codes used for company parts agree in principle with those used on AN, MS, and AS engine and propeller standard utility parts except that the letter "E" prefixing the standard utility parts material code has been omitted on titanium alloy materials, corrosion resistant materials, and heat resistant materials. The prefix letter "E" was replaced with prefix letter "S" for common temperature range materials.

3.4 Material codes have been assigned for common temperature range, corrosion resistant, heat resistant, and titanium materials as shown in Figure 1.

4. NEW MATERIALS:

Upon request the SAE will consider assigning material codes to materials not listed in this document. The SAE will coordinate proposed additions or alterations in this document with SAE Committee E-25. Material codes should not be assigned to new materials without the approval of the SAE.

5. CANCELLED MATERIAL CODES:

The following documents and their respective material codes have been cancelled:

E25	S25	AMS 6353
E29	S29	AMS 6380
EH10	H10	AMS 5767

6. NOTES:

(a) Inactive for new design after September 1, 1959. Materials of the same type (i.e., same chemical composition) have been reassigned a single code regardless of raw stockform.

(b) The following material specifications are "Noncurrent":

AMS 5750
AMS 6418
AMS 6485
AMS 6535
AMS 6550

(c) The following material specifications have been cancelled and replaced by an "AS" specification.

AMS 7225 replaced by AS7225
AMS 7228 replaced by AS7228
AMS 7229 replaced by AS7229
AMS 7232 replaced by AS7232